IN THE CLAIMS:

Please enter the following claims as amended:

1. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein the actuating mechanism comprises a shape memory material element and wherein the tabs are rotatably attached to the nozzle with the shape memory material element having a first end and a distal end, the actuation mechanism comprises the shape memory element attached at said first end to the nozzle and attached at said distal end to said tab, such that, in use, the element in a first shape maintains the tab in the second non-deployed position ~~wherein the actuation mechanism comprises a shape memory material element and wherein the nozzle further comprises a radially inner part and a radially outer part, wherein the tabs are rotatably attached to the nozzle at the radially inner part, the actuation mechanism comprises the shape memory element mounted at a first end to a radially outer part of the nozzle and mounted at a distal end said radially outer part of the tab, such that in use, the element in a first shape maintains the tab in the second non-deployed~~.

2. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 1 wherein the plurality of tabs is circumferentially disposed about the nozzle.

Claims 3 and 4 were previously cancelled.

5. (previously presented) A gas turbine engine exhaust nozzle arrangement as claimed in claim 1 wherein the periphery of the nozzle defines a pocket therein

and at least a part of the element is generally disposed within the pocket.

6. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interact with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs are substantially aerodynamically unobtrusive wherein the actuation mechanism comprises a shape memory material element and wherein the tab defines a recess therein and at least a part of the element is generally disposed within the recess.

7. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 1 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the actuation mechanism comprises a shape memory material element and wherein the element is ~~in the form of~~ a spring.

8. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise

thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein the nozzle arrangement comprises a resilient member having a first end and a distal end, the resilient member is attached at the first end to the tab and at the distal end to the nozzle and is arranged to provide a returning force to the tab.

9. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein the actuation mechanism comprises a shape memory material element and wherein the nozzle defines an orifice and a passage, the orifice is exposed to a gas stream and the passage extends from the orifice to the pocket and thereby provides a conduit for transmitting the thermal flux of the gas stream to the actuation mechanism.

10. (currently amended) A gas turbine engine exhaust nozzle arrangement ~~as claimed in claim 1~~ for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive and wherein the tab comprises shape memory material.

11. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 wherein the tab further comprises a flexural element, the flexural

element, in use, is arranged to provide a returning force to the tab.

12. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 wherein the tab defines an orifice, the orifice exposed to a gas stream, and a passage, the passage extending from the orifice, to the shape memory material and thereby provides a conduit for rapidly transmitting changes in the thermal flux of the gas stream to and throughout the memory shape material element.

13. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 1 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the actuation mechanism is actuated in a response to an applied field.

14. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 13 wherein the field is a temperature flux.

15. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 13 wherein the field is an electric current.

16. (currently amended) A gas turbine engine exhaust nozzle arrangement ~~as claimed in claim 13~~ for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein the actuation mechanism is actuated in a response to an applied field and

wherein the temperature flux is provided by the gas stream and the gas stream is any one chosen from the group comprising an ambient gas flow, a bypass flow, a core flow.

17. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[2]] 10 wherein the shape memory material element comprises any one of a group comprising Titanium, Manganese, Iron, Aluminium, Silicon, Nickel, Copper, Zinc, Silver, Cadmium, Indium, Tin, Lead, Thallium, Platinum.

18. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 2 wherein the shape memory material element comprises an electrostrictive material.

19. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 18 wherein the actuation mechanism further comprises an electrical circuit, the electrical circuit comprising control apparatus, an electric generating means and electrical contact means, the electrical contact means arranged to deliver, in use, an electrical signal, generated by the electrical generating means, through the electrostrictive material, the control apparatus operable to control the electrical signal.

20. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 19 wherein when the control apparatus is operated to deliver the electrical signal to the electrostrictive material, thereby actuating the electrostrictive material, the tab is moved from a second non-deployed position and a first deployed position and when the control means is operated so as not to deliver the electrical signal the electrostrictive material moves the tab between the first deployed position and the second non-deployed position.

21. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 19 wherein when the control apparatus is operated to deliver the electrical signal to the electrostrictive material, thereby actuating the electrostrictive material, the tab is moved between a first deployed position and a second non-deployed position and when the control means is operated so as not to deliver the electrical signal the electrostrictive material moves the tab from the second non-deployed position ~~and~~ to the first deployed position.

22. (currently amended) A gas turbine engine exhaust nozzle arrangement ~~as claimed in claim 19~~ for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interact with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs are substantially aerodynamically unobtrusive wherein the plurality of tabs is circumferentially disposed about the nozzle wherein the shape memory material element comprises an electrostrictive material wherein the actuation mechanism further comprises an electrical circuit, the electrical circuit comprising control apparatus, an electric generating means and electrical contact means, the electrical contact means arranged to deliver, in use, an electrical signal, generated by the electrical generating means, through the electrostrictive material, the control apparatus operable to control the electrical signal. wherein the control apparatus, operable to control the electrical signal and is operated in response to the altitude of an associated aircraft.

23. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 18 wherein the electrostrictive material element comprises any one of a group comprising Lead Zirconate Titanate, Lead Magnesium Niobate and Strontium Titanate.

24. (currently amended) A gas turbine engine exhaust nozzle arrangement ~~as claimed in claim 18~~ for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interact with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs are substantially aerodynamically unobtrusive wherein the plurality of tabs is

circumferentially disposed about the nozzle wherein the shape memory material element comprises an electrostrictive material and wherein the electrostrictive material element comprises any one of a polymer group including polyvinylidene fluoride.

25. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the downstream portion of the nozzle comprises a downstream periphery, the plurality of circumferentially disposed tabs extending in a generally downstream direction from the downstream periphery.

26. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein the downstream portion of the nozzle defines a plurality of circumferentially disposed recesses, each recess receiving a tab.

27. (previously presented) A gas turbine engine exhaust nozzle arrangement as claimed in claim 26 wherein said tab, in said second non-deployed position, substantially occupies said recess.

28. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 1 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the

tabs comprise a thermal barrier coating disposed ~~to~~ on a surface thereof.

29. (currently amended) 29. (amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 1 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the nozzle comprises a thermal barrier coating disposed to a surface thereof.

30. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the tabs circumferentially taper in the downstream direction.

31. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein said nozzle includes a nozzle wall and the tabs are radially inwardly angled at an angle of up to 20° relative to the nozzle wall.

32. (currently amended) A gas turbine engine exhaust nozzle as claimed in claim 10 ~~arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein said nozzle includes a nozzle wall and the tabs are radially outwardly angled at an angle of up to 20° relative to the nozzle wall.

33. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 ~~for the flow of exhaust gases therethrough between an~~

~~upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein said nozzle includes a nozzle wall and the tabs are circumferentially alternately radially inwardly angled at an angle of up to 20° relative to the nozzle wall and radially outwardly angled at an angle of up to 20° relative to the nozzle wall.

34. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the tabs are of a substantially trapezoidal shape.

35. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the general shape of the tabs is any one of the group comprising rectangular, square and triangular shape.

36. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein said nozzle includes a nozzle wall and the tabs are circumferentially disposed about the periphery of the nozzle wall to define substantially trapezoidal shaped notches between adjacent tabs.

37. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein said nozzle includes a nozzle wall and the tabs are circumferentially disposed about the periphery of the nozzle wall to define substantially V-shaped notches between adjacent tabs.

38. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the edges of the tabs are curved.

39. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein said nozzle includes a nozzle all and the nozzle tabs are radially inwardly angled at an angle of up to 10° relative to the nozzle wall.

40. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the tabs extend in circumferentially alternating radially inward and outward directions

for mixing the exhaust gas streams.

41.(currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 2 wherein the actuation mechanism comprises a shape memory material element spanning between each circumferentially adjacent ~~deployable~~ tab, the shape memory material element having a first length and a second length, so that in use, when the shape memory material element is in its first shape the ~~deployable~~ tabs are in the first deployed position and when the shape memory material element is in its second shape the ~~deployable~~ tabs are in the second non-deployed position.

42. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein the plurality of tabs is circumferentially disposed about the nozzle wherein the actuation mechanism comprises a shape memory material element spanning between each circumferentially adjacent deployable tab, the shape memory material element having a first length and a second length, so that in use, when the shape memory material element is in its first shape the deployable tabs are in the first deployed position and when the shape memory material element is in its second shape the deployable tabs are in the second non-deployed position wherein the first length of the shape memory element is longer than the second length, so that in use and in the first deployed position the deployable tabs are angled radially outwardly.

43. (original) A gas turbine engine exhaust nozzle arrangement as claimed in claim 41 wherein the first length of the shape memory element is shorter than the second length, so that in use and in the first deployed position the deployable

tabs are angled radially inwardly.

44. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein, in use as a noise reduction means, alternate tabs are rigidly fixed at a radially inward angle and deployable tabs are operable to move between said first deployed position at a radially outward angle, where the deployable tabs interact with a gas stream to reduce exhaust noise thereof, and said second non-deployed position, where the deployable tabs are substantially circumferentially aligned with the alternate tabs.

45. (currently amended) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab ~~extends~~ extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive wherein, in use as a noise reduction means, alternate tabs are rigidly fixed at a second non-deployed position and deployable tabs are operable to move between [[a]] said first deployed position at a radially inward angle, where the deployable tabs interact with a gas stream to reduce exhaust noise thereof, and [[a]] said second non-deployed position, where the deployable tabs are substantially circumferentially aligned with the alternate tabs.

46. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 1 wherein said nozzle has a downstream periphery and the downstream periphery comprises straight edges, each straight edge having a tab disposed thereto.

47. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the actuation mechanism further comprises an end stop, the end stop is configured to provide a positive locator for the tab in either its deployed or non-deployed positions.

48. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the exhaust nozzle is a core engine nozzle.

49. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the exhaust nozzle is a bypass exhaust nozzle.

50. (currently amended) A ducted fan gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the arrangement comprises a core exhaust nozzle and a bypass exhaust nozzle.

51. (currently amended) A ducted fan gas turbine engine exhaust nozzle arrangement comprising an outer bypass exhaust nozzle as claimed in claim [[1]] 10.

52. (currently amended) A ducted fan gas turbine engine exhaust nozzle arrangement as claimed in claim 51 wherein said exhaust nozzle has a downstream periphery and said bypass nozzle has a downstream end and said [[the]] downstream end of the bypass nozzle is further downstream than [[the]] said downstream periphery of the core exhaust nozzle.

53. (original) A ducted fan gas turbine engine exhaust nozzle arrangement as claimed in claim 50 wherein the downstream end of the bypass nozzle is upstream of the downstream periphery of the core exhaust nozzle.

54. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the arrangement is for exhaust noise attenuation.

55. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim [[1]] 10 wherein the tabs extend generally in a downstream direction.

56. (currently amended) A gas turbine engine exhaust nozzle arrangement as claimed in claim 10 ~~for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extends in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving the tabs between a first deployed position, in the first position the tabs interacting with a gas stream to reduce exhaust noise thereof, and a second non-deployed position, in the second position the tabs being substantially aerodynamically unobtrusive~~ wherein the tabs extend generally in an upstream direction.

Claim 57 was previously cancelled.

58. (new) A gas turbine engine exhaust nozzle arrangement for the flow of exhaust gases therethrough between an upstream end and a downstream end thereof comprising a nozzle, a downstream portion and a plurality of tabs, each tab extending in a generally axial direction from the downstream portion of the nozzle wherein the nozzle further comprises an actuation mechanism capable of moving at least some of the tabs between a first deployed position, in the first position the tabs being angled radially outward relative to the nozzle and interact with a gas stream to reduce exhaust noise thereof and to increase the area of the nozzle, and a second position, in the second position the tabs reducing the nozzle area relative to the first position wherein the tab comprises shape memory material.

59. (new) A gas turbine engine exhaust nozzle arrangement as claimed in claim 58 wherein others of the tabs are fixed in the second position.

60. (new) A gas turbine engine exhaust nozzle arrangement as claimed in claim 59 wherein circumferentially adjacent tabs interlock with one another when in the second position.

61. (new) The gas turbine engine exhaust nozzle arrangement as claimed in claim 59 wherein circumferentially adjacent tabs overlap one another when in the second position.

62. (new) A gas turbine engine exhaust nozzle arrangement as claimed in claim 58 wherein other tabs are fixed in the second position and interact with a gas stream to reduce exhaust noise thereof when the said at least some of the tabs are in the first position